PAB3D simulations for 3rd AIAA Drag Prediction Workshop

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Presentation Outline

- Introduction
- Governing Equations
- Results and Discussion
- Summary

Introduction

- Case 1 DLR-F6-WB and FX2B configurations
- Code PAB3D structured multi-block
- Grid Boeing grids
  (coarse, medium, medium-fine, & fine)
- Mach = 0.75, Re = 5 million

Methodology
PAB3D CFD Code

- 3-D RANS Upwind Code
- Multi-block structured with general patching
- Parallel computing using MPI
- Mesh sequencing
- Low memory requirement
- Graphical user interface (GUI) capabilities
- Linear two-equation k-ε models
- Suite of algebraic Reynolds stress models
- LES & PANS

Governing Equations

Reynolds (Favre) Averaged Navier-Stokes

RANS

Continuity

$$\frac{\partial \rho}{\partial t} + \frac{\partial \rho u_i}{\partial x_i} = 0$$

Momentum

$$\frac{\partial \rho u_i}{\partial t} + \frac{\partial (\rho u_i u_j + p\delta_{ij})}{\partial x_j} = \frac{\partial (\tau_{ij} + \rho \overline{u_i u_j})}{\partial x_j}$$

Energy

$$\frac{\partial \rho e_0}{\partial t} + \frac{\partial (\rho e_0 u_i + p u_i)}{\partial x_i} = \frac{\partial (\tau_{ij} u_j - \rho \overline{u_i u_j} u_j)}{\partial x_i}$$

$$- \frac{\partial (q_i - C_P \rho \overline{u_i t})}{\partial x_i} - \frac{\rho \overline{u_i u_j u_j}}{2}$$

Reynolds Stress

Governing Equations

Two Equation K- ε Model

$$\frac{\partial \rho k}{\partial t}+\frac{\partial \rho u_i k}{\partial x_i}=-\rho\overline{u_i u_j}\frac{\partial u_j}{\partial x_i}+\frac{\partial}{\partial x_i}\left[\mu_l+\frac{c_\mu k^2}{\sigma_k \varepsilon}\frac{\partial k}{\partial x_i}\right]-\rho\varepsilon(1.+M_\tau^2)$$

$$\frac{\partial \rho \varepsilon}{\partial t}+\frac{\partial \rho u_i \varepsilon}{\partial x_i}=-C_{\varepsilon 1}\rho\overline{u_i u_j}\frac{\partial u_j}{\partial x_i}\frac{\varepsilon}{k}+\frac{\partial}{\partial x_i}\left[\mu_l+\frac{c_\mu k^2}{\sigma_\varepsilon \varepsilon}\frac{\partial \varepsilon}{\partial x_i}\right]-f_2\tilde{C}_{\varepsilon 2}\rho\frac{\varepsilon}{k}\left[\varepsilon-\nu_l\left(\frac{\partial\sqrt{k}}{\partial n}\right)^2\right]$$

Reynolds Stress

$$C_\mu=.09,\ C_{\varepsilon 1}=1.44,\ and\ \tilde{C}_{\varepsilon 2}=C_{\varepsilon 2}=1.92$$

Turbulent viscosity: $$\upsilon_t^{RANS}=f_\mu \rho\, C_\mu \frac{k^2}{\varepsilon}$$

Governing Equations

Turbulent Stresses

Two Equation K- ε Linear Model
(NASA CR-4702, December 1995)

$$-\overline{u_i u_j} = 2\upsilon_t S_{ij} - \frac{2}{3}\delta_{ij} k$$

Shih, Zhu, & Lumley (SZL) Nonlinear Model
NASA TM-106644, August 1994

$$-\overline{u_i u_j} = 2\upsilon_t S_{ij} - \frac{2}{3}\delta_{ij} k + 2\beta \frac{K^3}{\varepsilon^2}(W_{ik}\bar{S}_{kj} - \bar{S}_{ik}W_{kj})$$

Girimaji Nonlinear Model
ICASE 95-82, December 1995

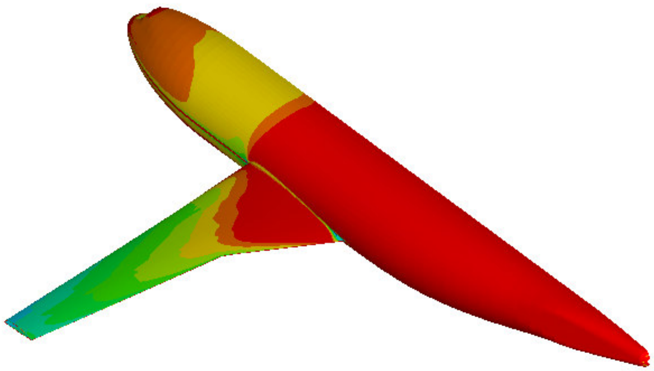
$$-\overline{u_i u_j} = 2\upsilon_t S_{ij} - \frac{2}{3}\delta_{ij} k + 2C_\mu^* \frac{K^3}{\varepsilon^2}[-G_2(W_{ik}S_{kj} - S_{ik}W_{kj}) + G_3(S_{ik}S_{kj} - \frac{1}{3}S_{mn}S_{mn}\delta_{ij})]$$

where

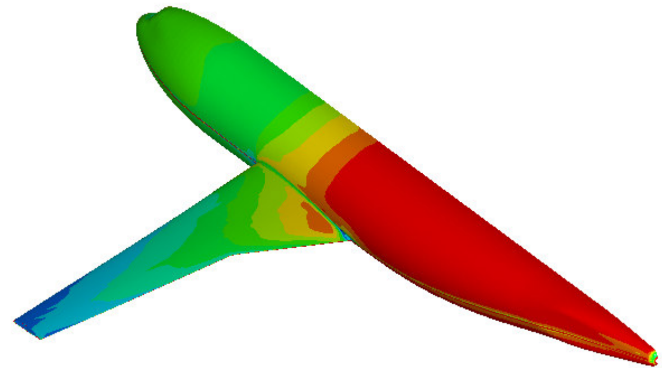
$$W_{ij} = \frac{1}{2}\left(\frac{\partial u_i}{\partial x_j} - \frac{\partial u_j}{\partial x_i}\right) \ \& \ \bar{S}_{ij} = S_{ij} - \frac{1}{3}S_{kk}\delta_{ij}$$

Computational Grids, Y+

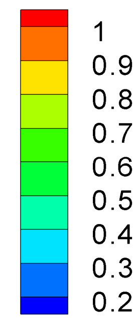
BOEING H-Grids generated by : T.J. Kao and N. J. Yu for Ed. Tinoco

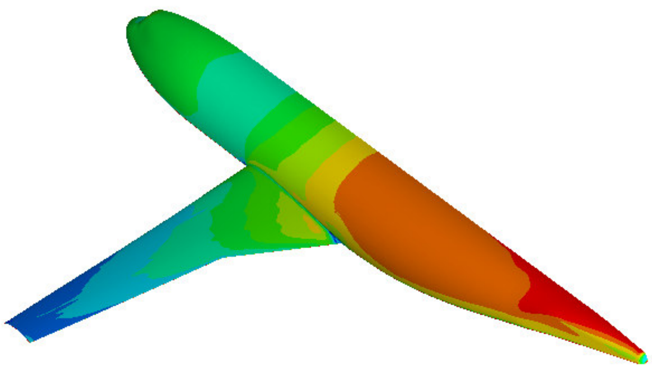


Coarse Grid
26 Blocks & 2298880 Cells

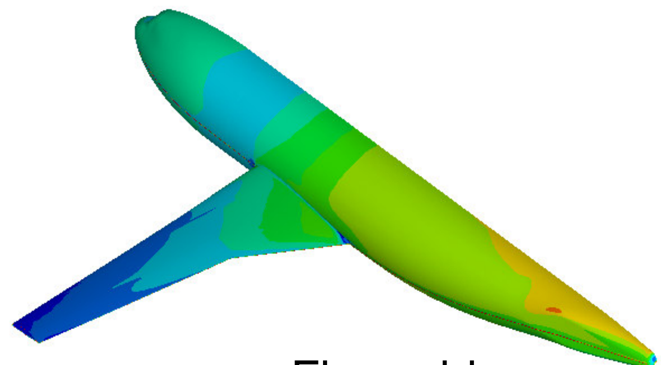


Medium grid
38 Blocks & 8080896 Cells





Medium-fine grid
26 Blocks & 15856320 Cells

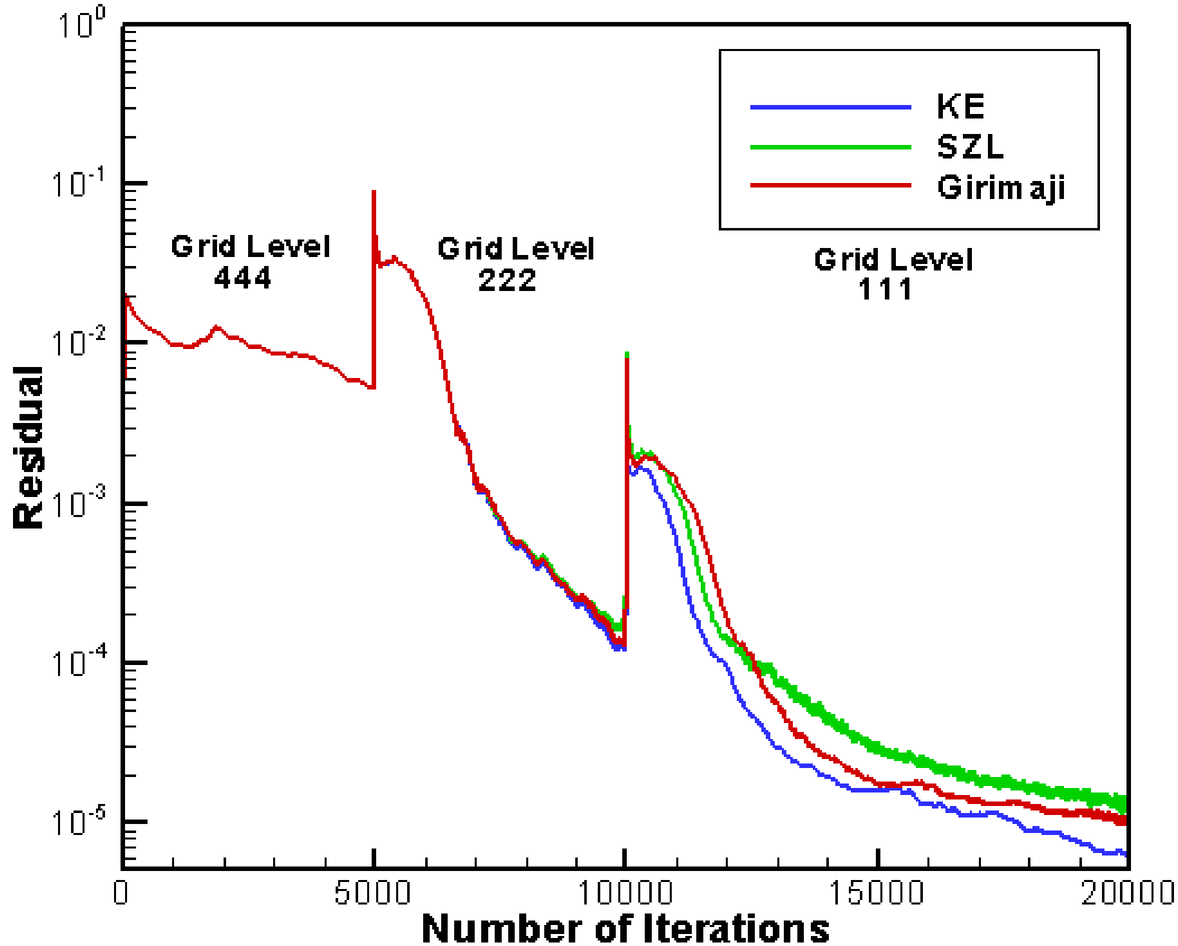


Fine grid
57 Blocks & 27185664 Cells

Case1, F6-WB & FX2B Mach 0.75, $Re=5*10^6$

Turbulence Model/ Grid	Coarse	Medium	Medium fine	Fine
k ε	X	X	X	X
SZL	X	X	X	X
Girimaji	X	X	X	X

Convergence History



Typical CPU Timings

Grid	Blocks	# of Cell * 10^6	Processors	Wall Time	Wall Time /grid point 10^{-6} sec
Coarse	26	$2.3 * 10^6$	24	02:02	0.4196472
Medium	38	8.0	38	5.42	0.3032175
Medium fine	49	15.8	46	8:33	0.2713972
Fine	58	27.2	56	17:45	0.2537843

Grid Convergence, Girimaji

Mach =0.75, Re=$5*10^6$

Grid	AOA	CL	CD	CM_TOT
Coarse	0.1	0.499	0.03163	-0.11880
Medium	0.132	0.501	0.02840	-0.13350
Medium fine	0.143	0.501	0.02731	-0.13590
Fine	0.143	0.500	0.02714	-0.13570

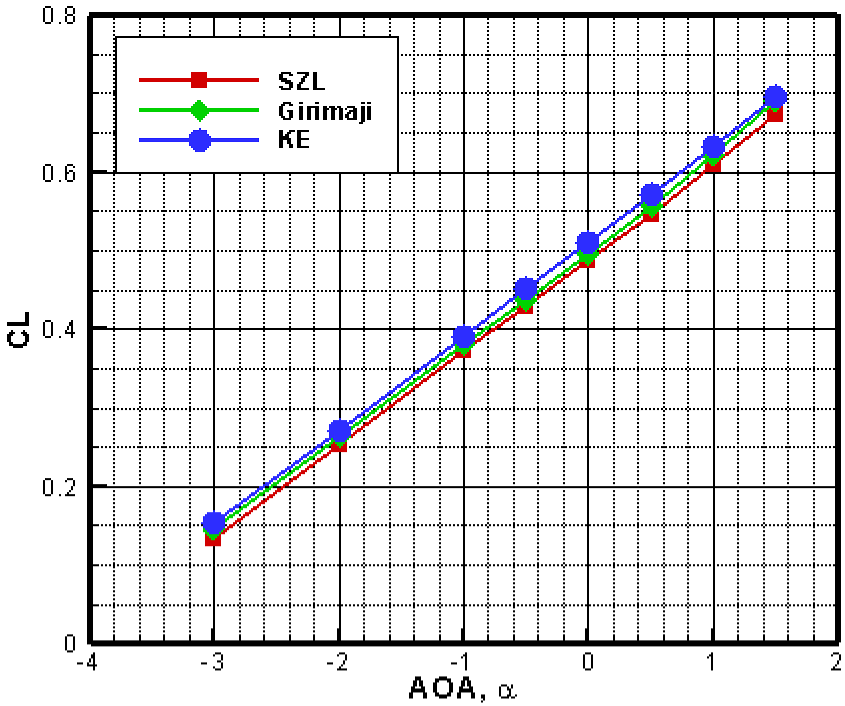
Turbulence Model Effect

Mach =0.75, Re=5*10^6

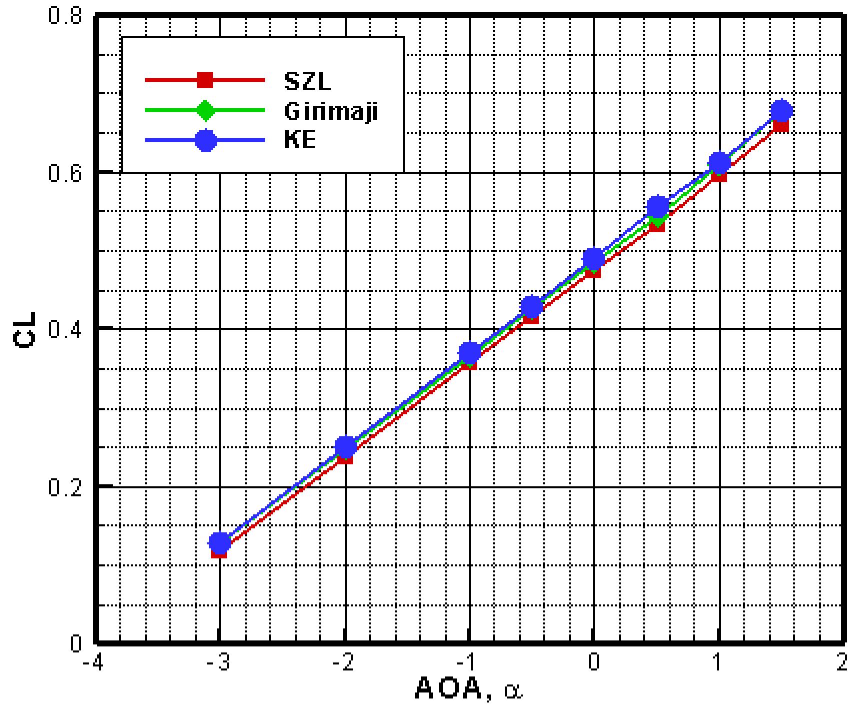
Grid	AOA	CL	CD	CM_TOT
KE	-0.067	**0.501**	0.03093	-0.1427
SZL	0.143	**0.500**	0.02714	-0.13570
Girimaji	**0.143**	**0.500**	0.02714	-0.13570

Lift Curve on Medium Grids

Mach =0.75, Re=5*10^6



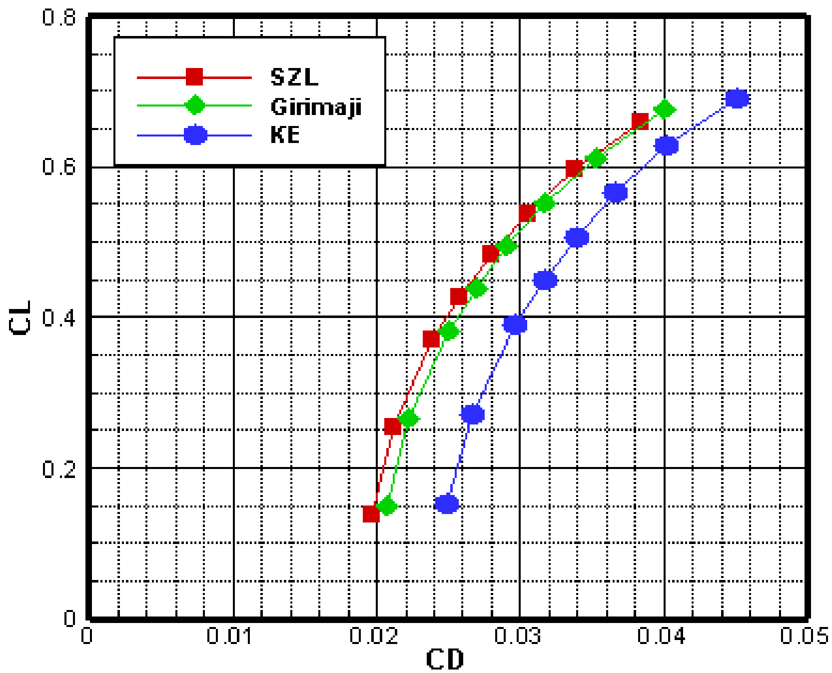
F6-WB



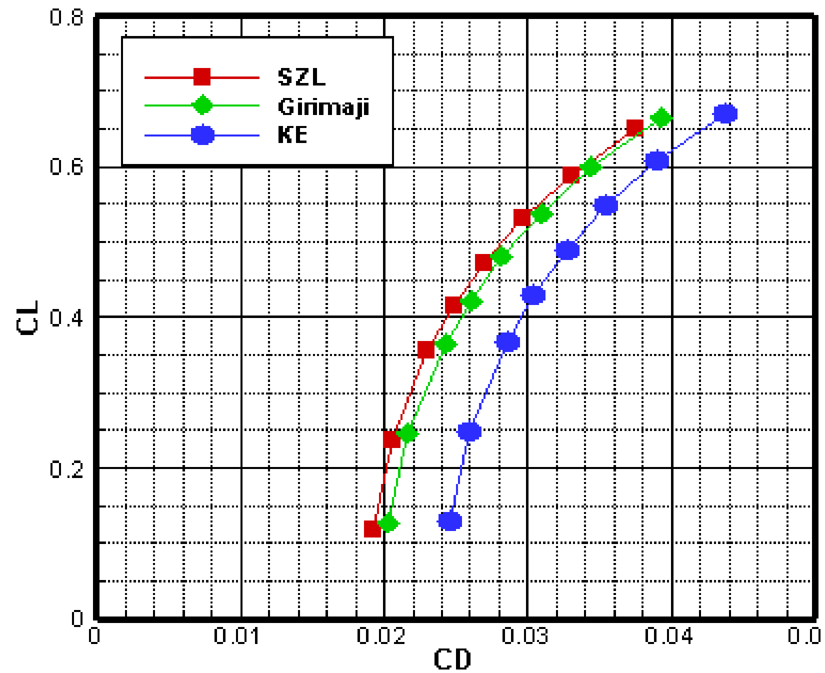
FX2B

Drag Polar on Medium Grids

Mach =0.75, Re=$5*10^6$

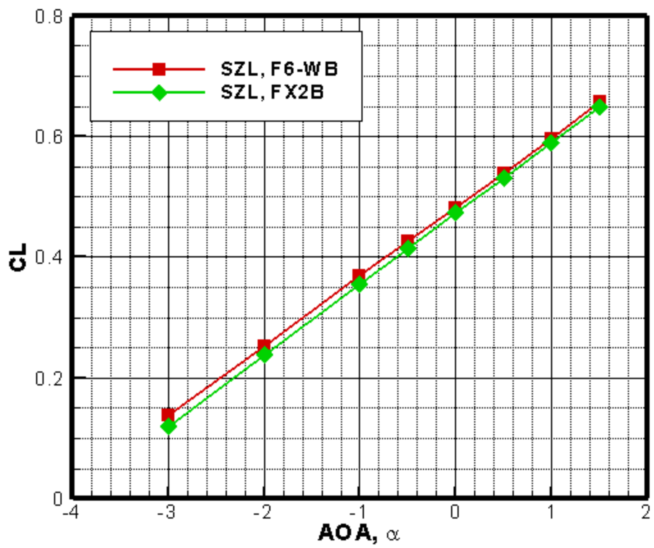


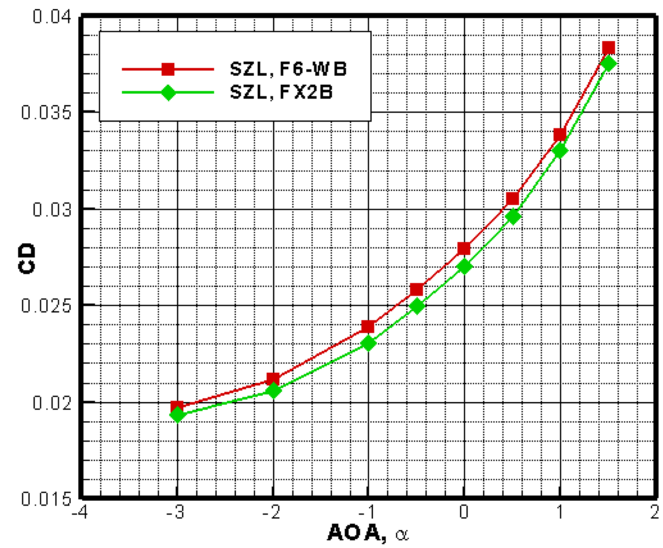
F6-WB

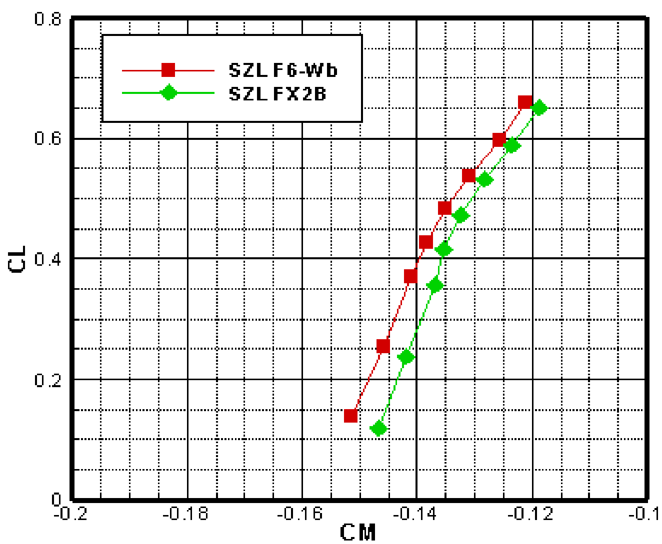


FX2B

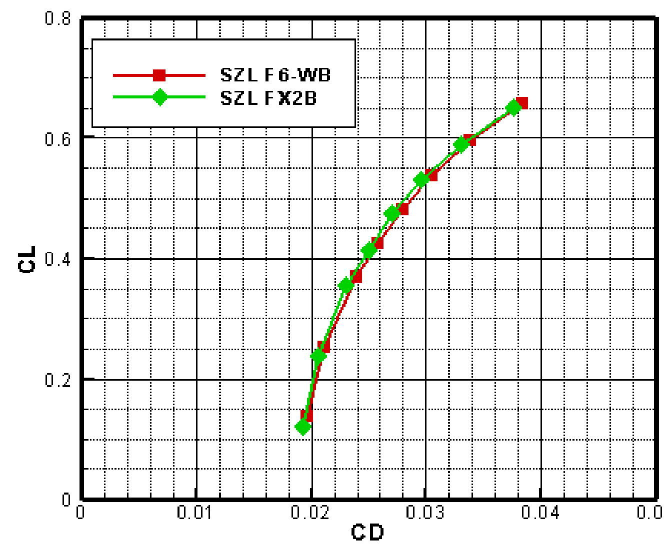
F6_WB and FX2B Comparison, SZL





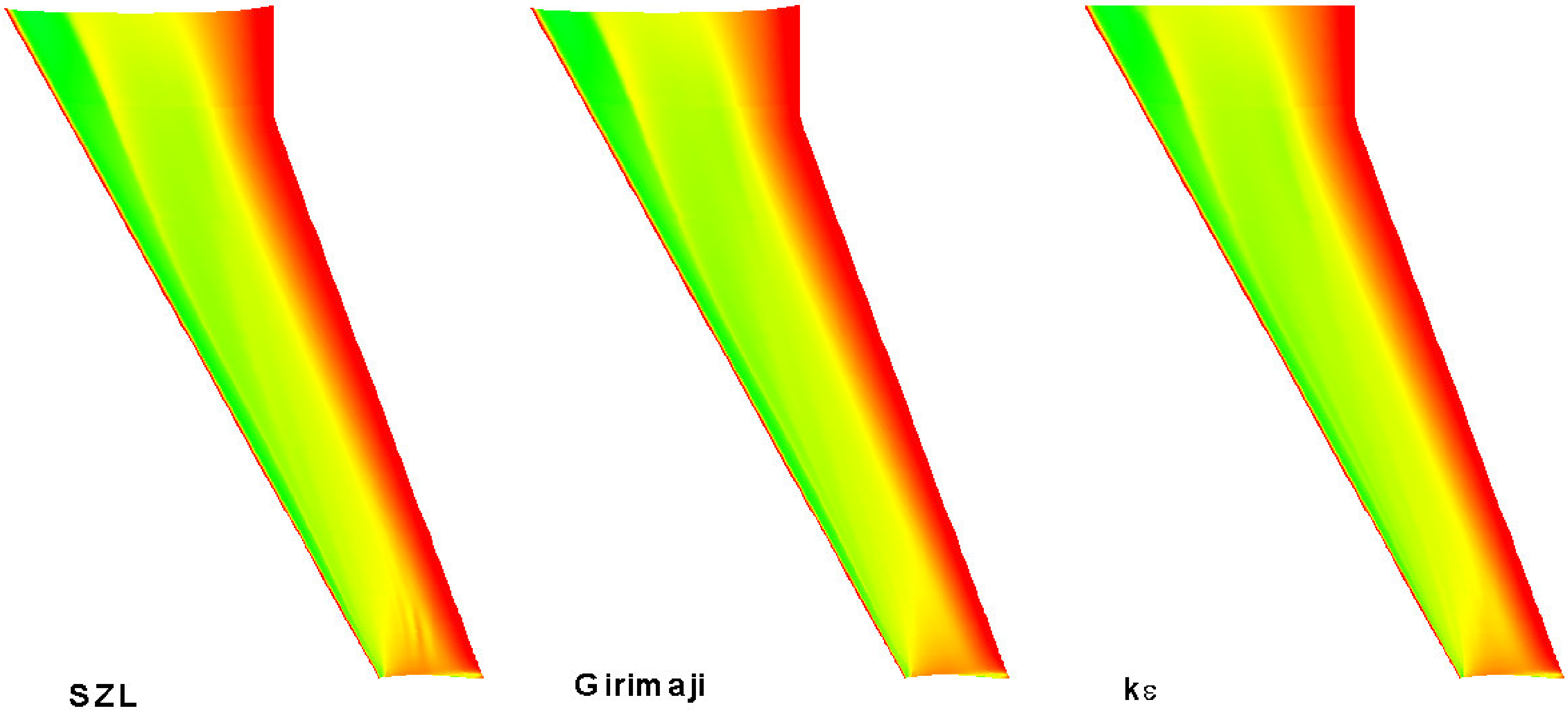


diction



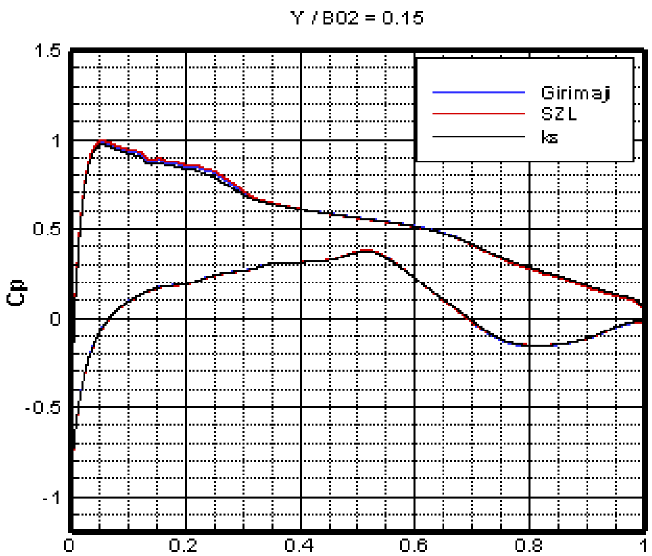
Coefficient of Pressure

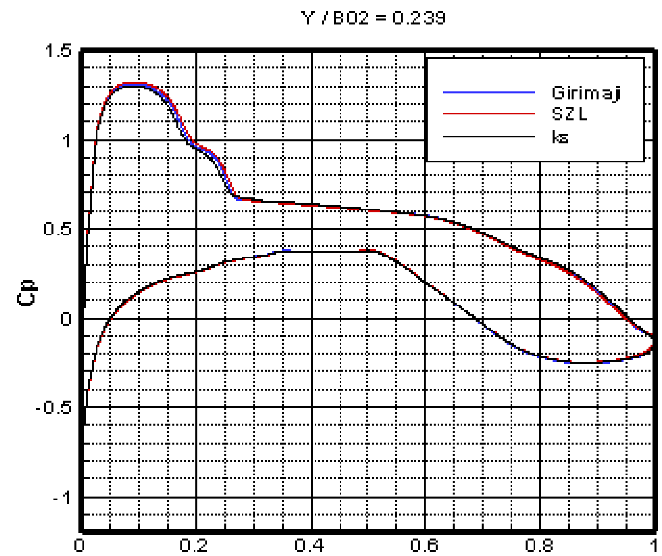
Mach =0.75, Re=5*10^6, AOA=0.0



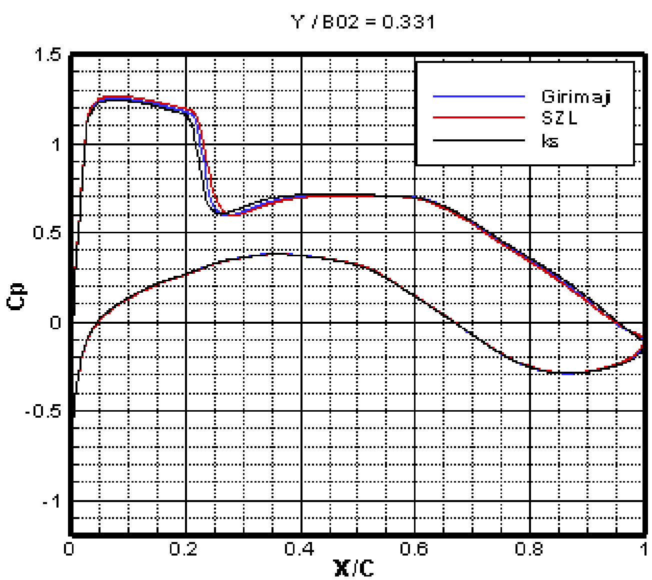
Turbulence Model Effect on Cp

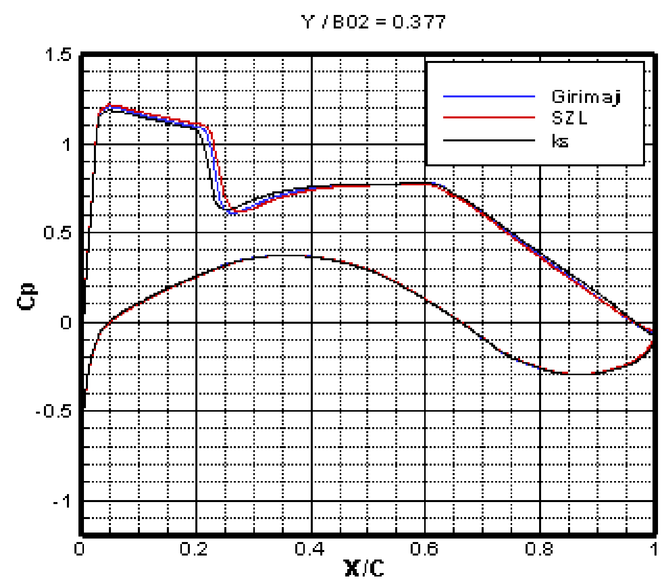
Mach =0.75, Re=5*10^6, CL=0.5





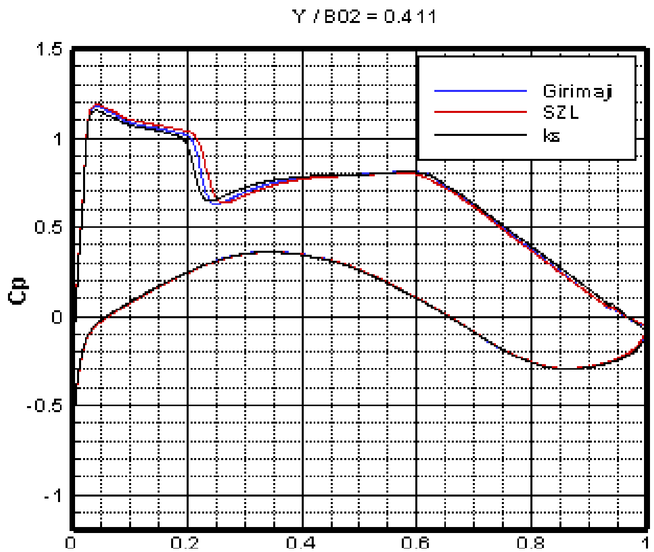
FX2B

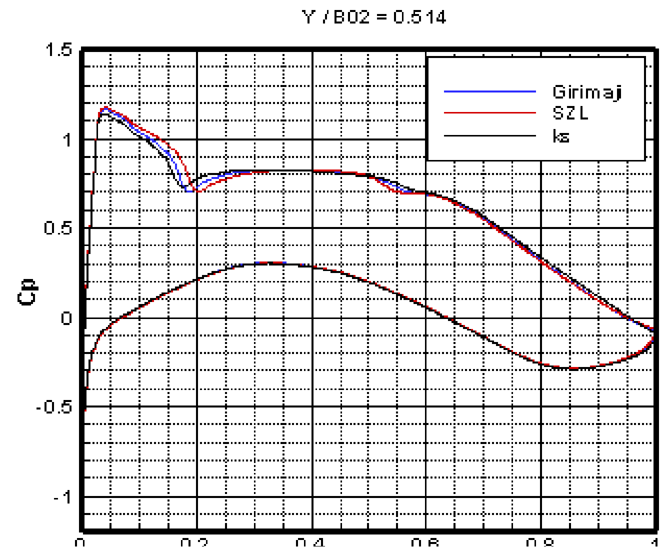




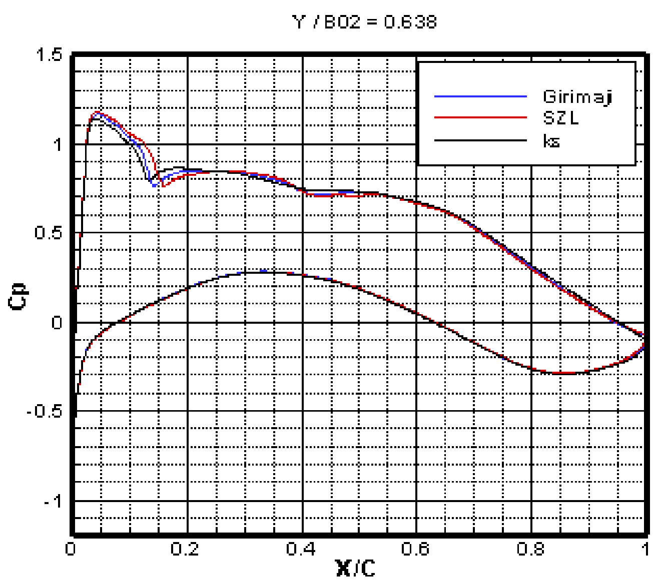
Turbulence Model Effect on Cp

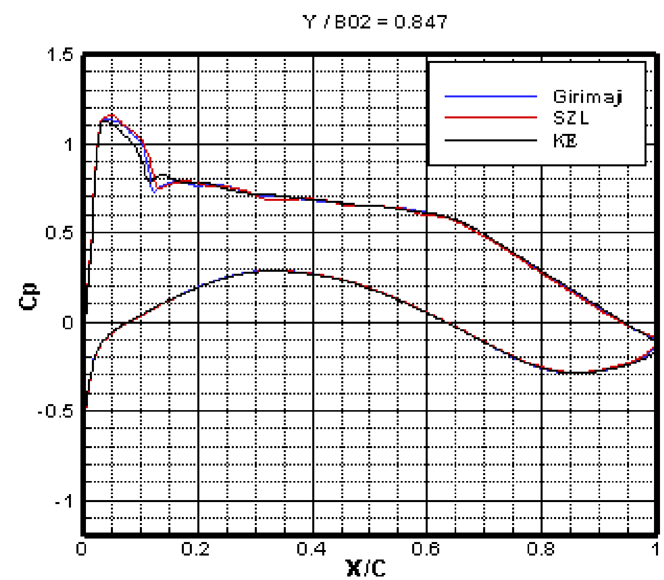
Mach =0.75, Re=5*10^6, CL=0.5





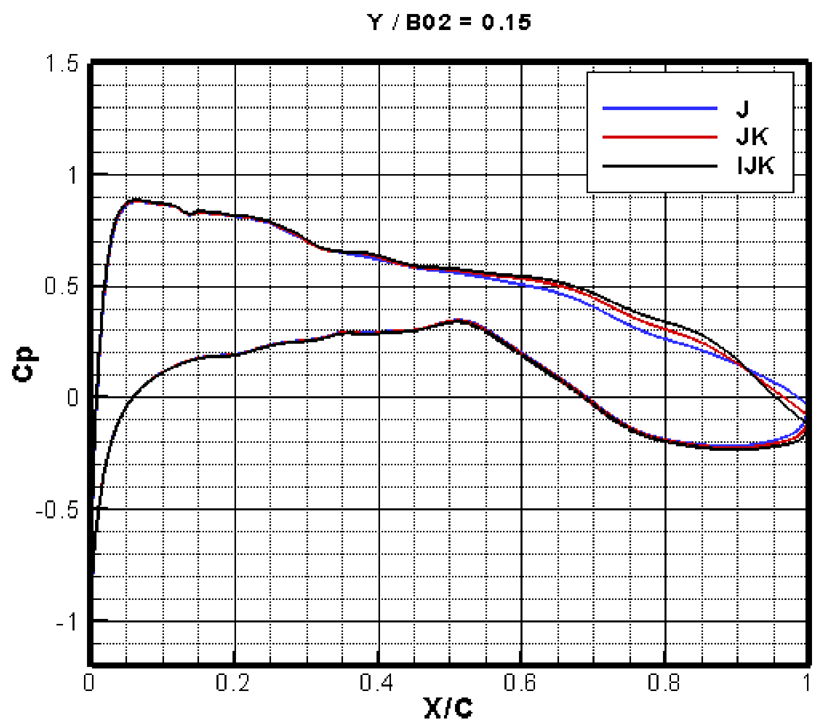
FX2B

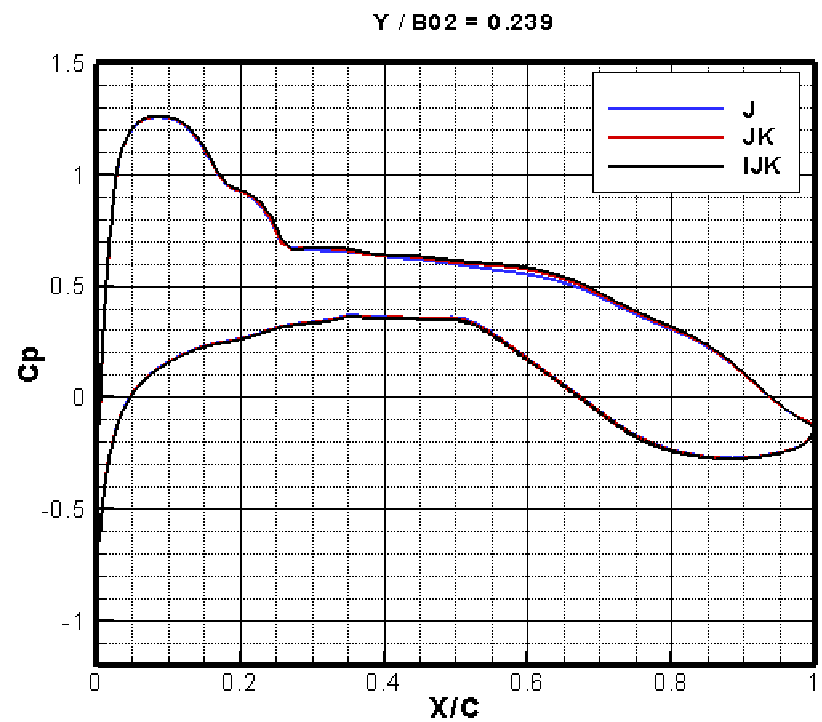




Thin Shear Layer Effect on Cp

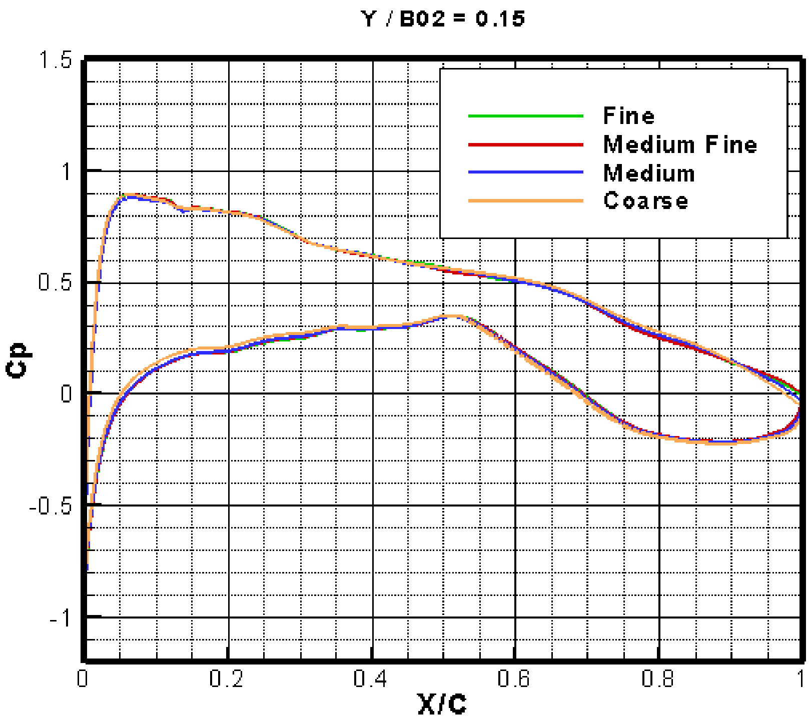
Mach =0.75, Re=5*10^6, AOA=0.0, SZL

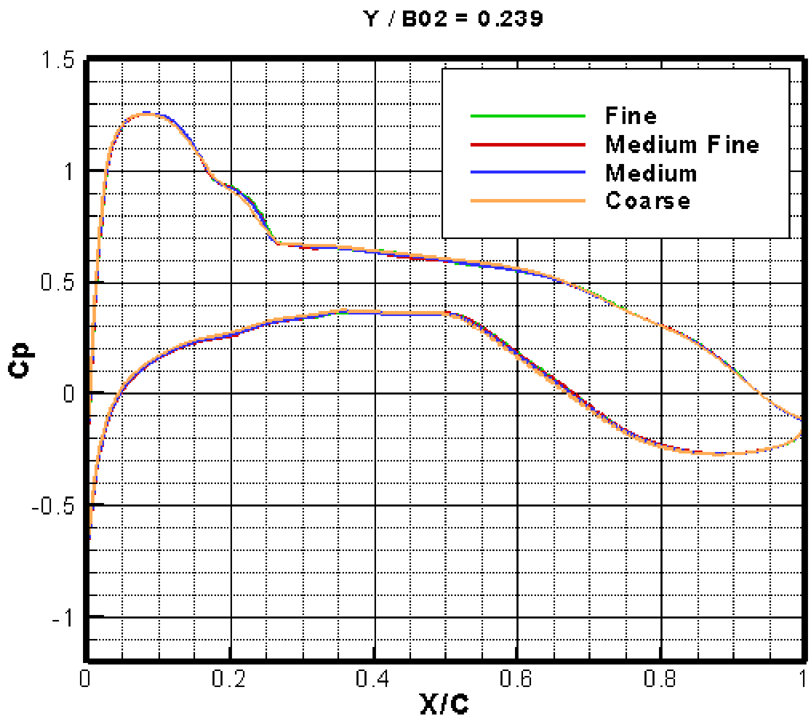




Grid Effect on Cp

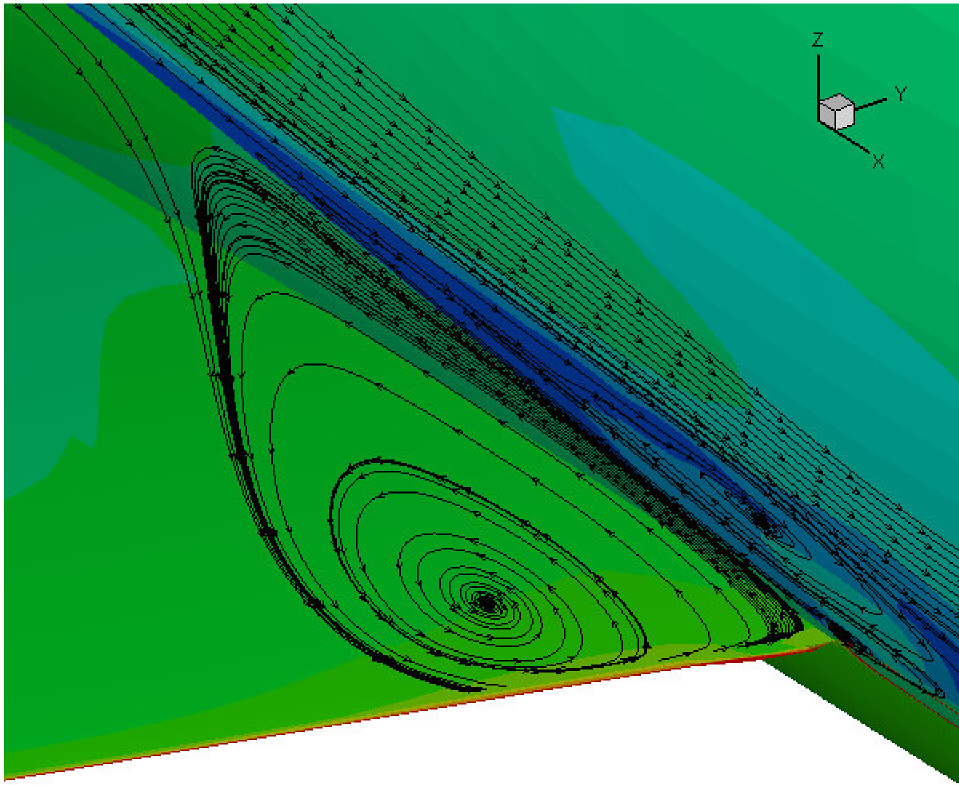
Mach =0.75, Re=5*10^6, AOA=0.0, SZL



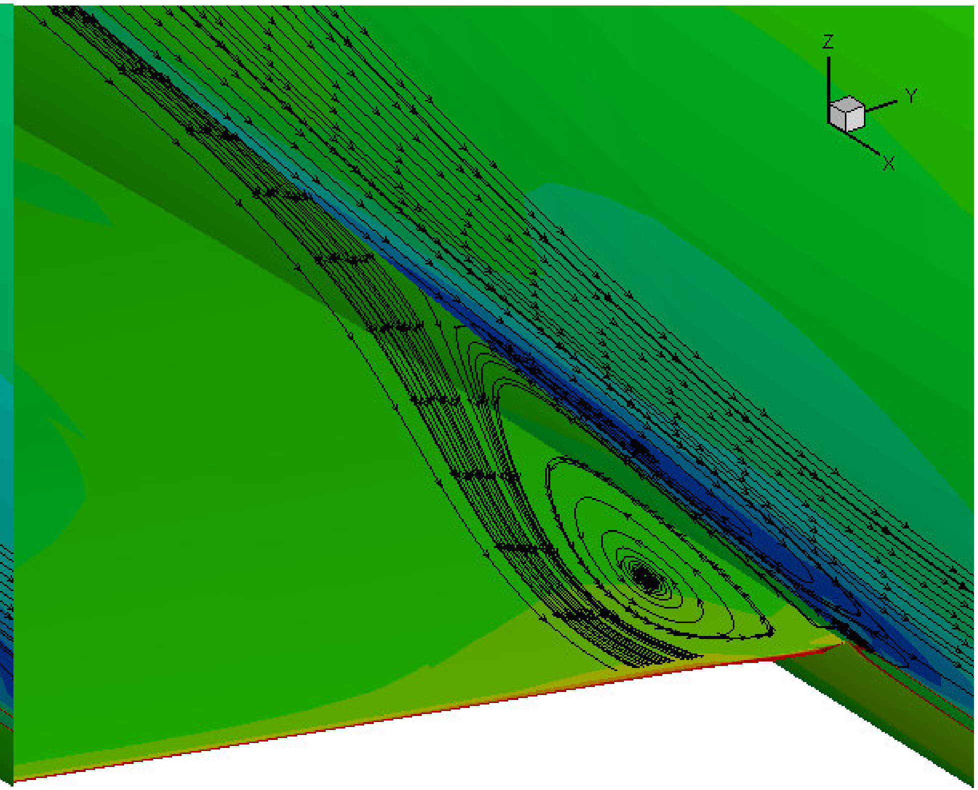


Upper Wing Streamlines, F6-WB

Mach =0.75, Re=5*10^6, AOA=0.0



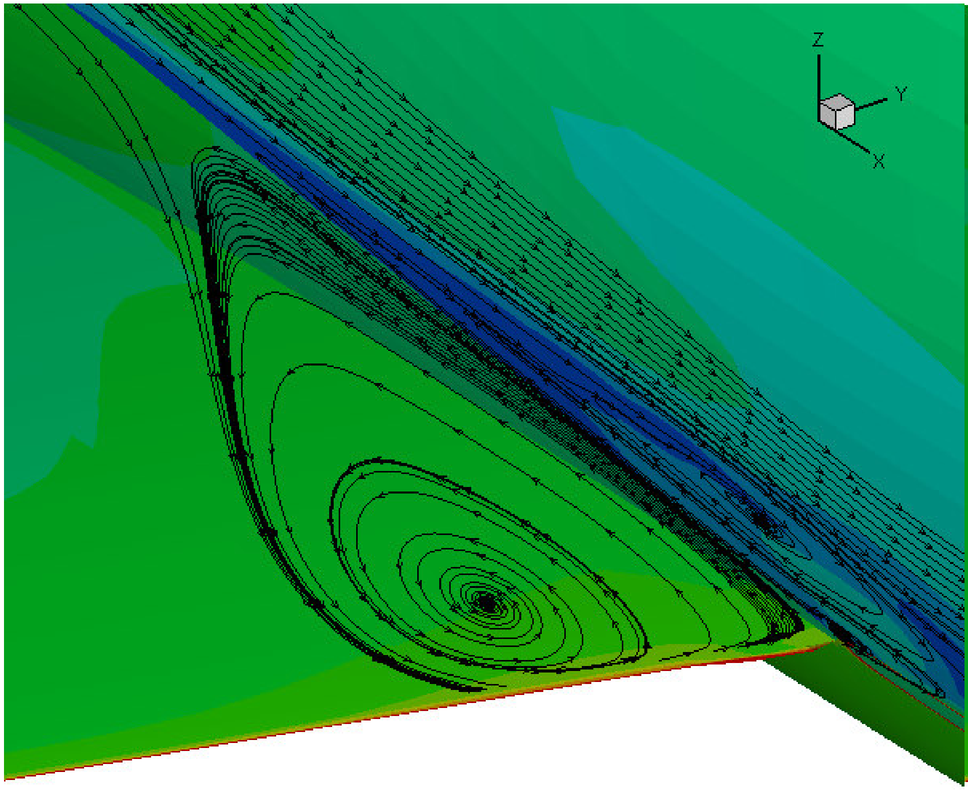
SZL medium grid
Thin Shear Layer in 2 directions



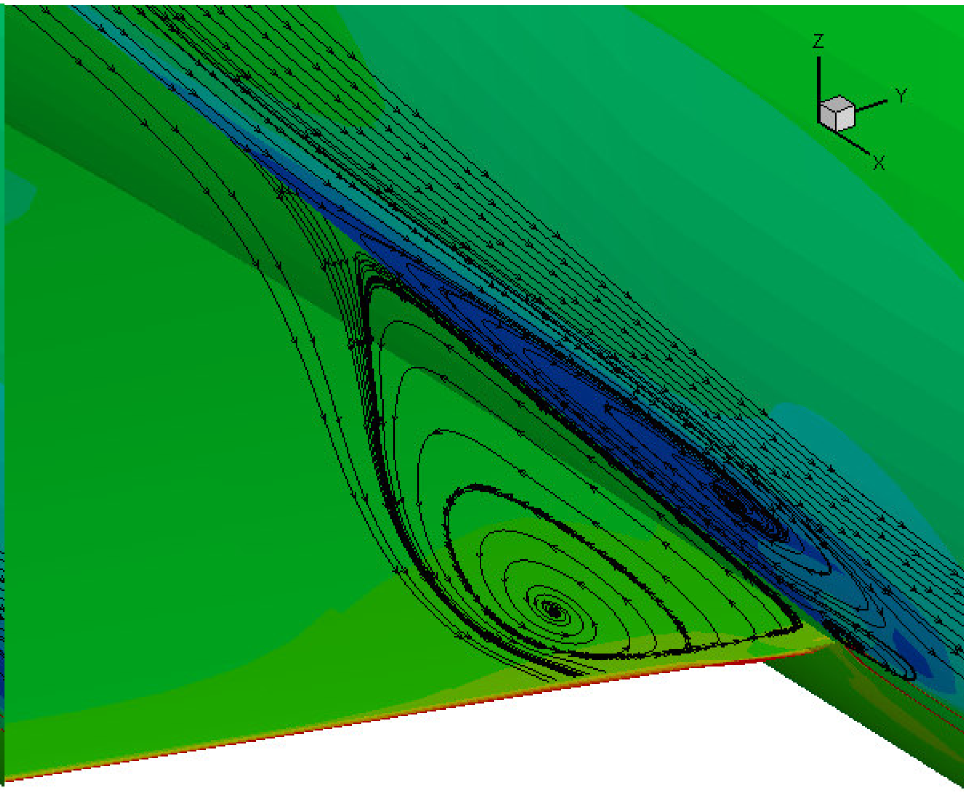
$k\varepsilon$ medium grid
Thin Shear Layer in 2 directions

Upper Wing Streamlines, F6-WB

Mach =0.75, Re=5*10^6, AOA=0.0



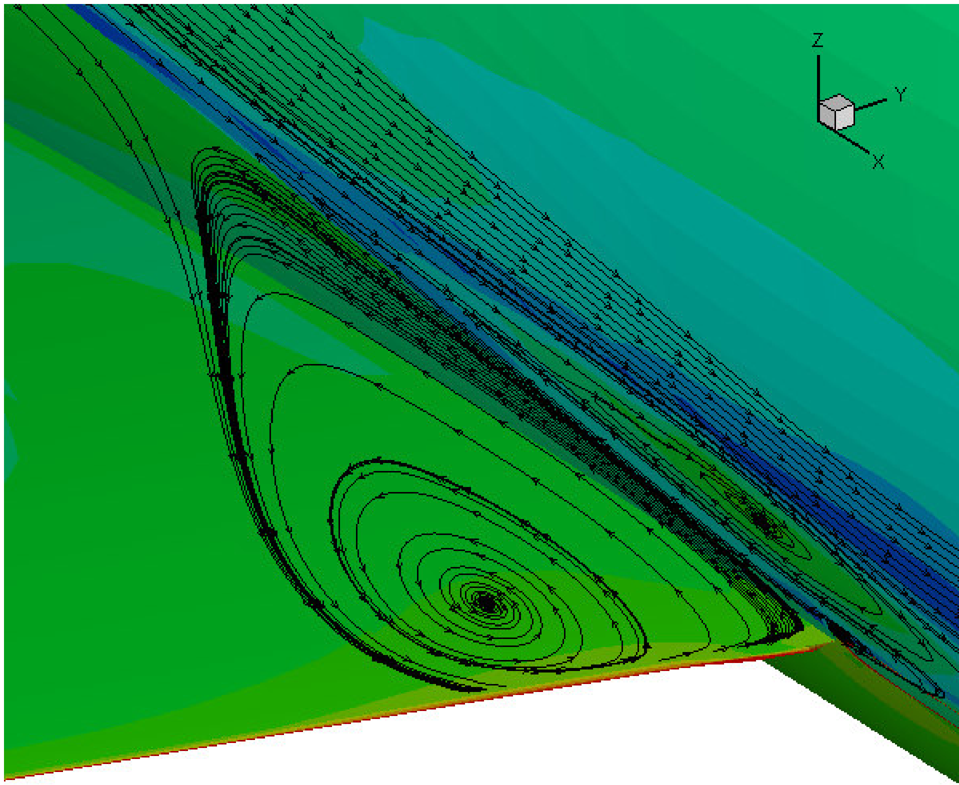
SZL medium grid



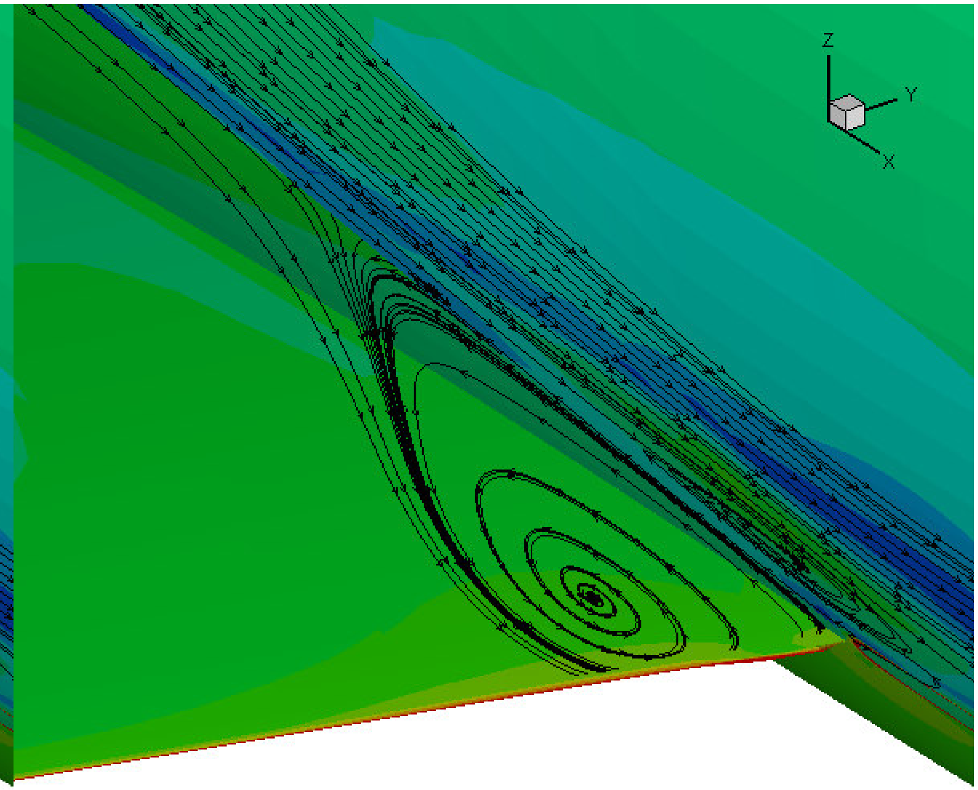
Girimaji medium grid

Upper Wing Streamlines, F6-WB

Mach =0.75, Re=$5*10^6$, AOA=0.0, SZL



Thin Shear Layer in 2 directions, JK



Thin Shear Layer in 3 directions, IJK

Summary

- PAB3D was utilized to compute the flow field of DLR F6-Wb & FX2B
- Three different turbulence Models and four different grid resolutions were utilized
- AS expected kε models produced the highest skin friction
- SZL & Girmaji algebraic Reynolds Stress models predicts similar separation bubbles
- Turbulence model affects size of separation bubble.
- Same suite of algebraic Reynolds Stress models are ported into USM3D

Need to develop innovative turbulence models for flow separation
and
implement the models within state-of-the-art RANS flow solvers